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ROBUST AERODYNAMIC OPTIMIZATION FOR HELICOPTER BLADE MORPHING AIRFOILS

Francesca Fusi[1], **Giuseppe Quaranta**[1], **Pietro M. Congedo**[2]

[1]Dipartiment of Aerospace Science and Technology
Politecnico di Milano
via La Masa, 34 Milano 20156 Italy
francesca.fusi@polimi.it
giuseppe.quaranta@polimi.it

[2]Team CARDAMOM – INRIA Bordeaux Sud-Ouest
200 Avenue de la Vieille Tour, Talence Cedex 33405 France
pietro.congedo@inria.fr

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Abstract: An aerodynamic optimization method is developed to define robust shapes for morphing airfoils for helicopter blades. The morphing strategy consists of a conformable camber airfoil which changes over the period of rotation of the blade to cope with the variable flow conditions encountered in forward flight. The paper is then focused on an aerodynamic optimization to set the optimal shape with respect to performance, where technological aspects are inserted through geometrical constraints. In fact, this paper presents an exploratory work on morphing geometries which aims at understanding the relationship between shape degree of freedom and actual aerodynamic gain. A robust or uncertainty-based approach is used to compute a reliable morphing airfoil, providing a low variance with respect to uncertainty affecting the operating conditions. In order to assess the effectiveness of the robust method, several optimization problems are performed, from a classical two-point drag minimization with lift coefficient constraint to a robust morphing camber optimization. The results of the optimization problems are compared and discussed to highlight the features of the robust approach. Exploring and demonstrating the gain of the aerodynamic shape may drive the development of new mechanism for the realization of morphing structures.

1 INTRODUCTION

Helicopter rotor blades present a real challenge to aerodynamic design. In fact, rotor blades are required to operate with satisfactory performance in extremely different conditions, depending on the azimuthal position and the advance ratio, and good performance in one condition does not transfer to another one. A trade-off solution must be pursued, whose performance in each condition over the flight envelope is sub-optimal. A very attractive solution to this problem is the employment of a blade capable of changing its shape during flight, that is a morphing blade. Morphing rotor design is a challenging and active area of research [1]: recent studies have focused on conformable airfoils to reduce vibration [2] or postpone dynamic stall onset [3]. Because morphing structures typically imply increased weight, complexity and cost, an in-depth assessment of the potential aerodynamic benefit of such an approach is strongly recommended [4]. Variable camber airfoils have been adopted in fixed-wing applications [5, 6]

and this solution has recently been explored for helicopter applications, as well. To this extent, a continuously changing rotor blade airfoil is seen as a successor to rotor blade trailing-edge flaps [2].

The flow conditions encountered by rotor blades may vary due to the effects of the blade dynamics and flexibility, as well as the interaction with vortices and wakes trailed from preceding blades 1. It follows that the flow conditions considered at the design stage for the definition of the optimal two-dimensional section are usually affected by uncertainty, not only because they are estimates of the actual condition, but also because modelling involves some level of approximation of the real flowfield. In such a context, taking into account the variability of the modelling parameters in the optimization procedure is a possible way to result in a more reliable optimal design. This is the goal of uncertainty-based optimization. Specifically, robust optimization seeks an optimal design that is minimally sensitive with respect to changes of the uncertain parameters that define the problem. The result is a robust design that maintains satisfactory performance also in off-design conditions. Few works consider uncertainty in the field of helicopter design [7–9]. In the application to morphing structures, a high level of uncertainty affects the modelling of the environmental conditions encountered by the airfoil, and therefore robustness becomes critical to define reliable design.

As a final remark, optimization of the morphing airfoil shape to improve the aerodynamic performance is typically run separately to the morphing mechanism design [10]. To this extent, the optimization approach is conceived as a two-level approach [5]: (i) the best deformed airfoil configuration is determined according to a desired aerodynamic performance, and (ii) in the second level, the best internal structural configuration is obtained using a topology optimization suited to the chosen compliant structure able to adapt itself in order to match the optimal shape coming out from the first level.

In this paper, a methodology for obtaining optimal morphing airfoils is presented, which considers different type of geometry constraints. The effect of camber and thickness morphing is discussed and trailing edge morphing is studied. The gain of the morphing airfoil strategy is assessed not only from a deterministic point of view, but also when considering uncertainty in the operating conditions. In such a way, a more realistic estimate of the gain of morphing strategies can be defined.

In the following, Section 2 is devoted to the description of some numerical ingredients, which are used in this paper. Then, a discussion on two different choices of morphing strategy is presented in Sec. 3 to assess the capability of morphing airfoils. In this first part, only deterministic evaluation are presented, because deterministic optimal shapes (and corresponding operating conditions) represent the good starting point of robust approaches. This discussion demonstrates how camber morphing is more effective to enhance the aerodynamic performance. Then, optimal shapes from deterministic and robust approaches are computed and presented in Sec. 4. Finally, the validation over the azimuthal angle is presented in Sec. 5. Some conclusions and perspectives are then drawn in Section 6.

2 NUMERICAL MODELS

2.1 Parameterization and Design variables

The Class/Shape function Transformation (CST) [11] is used to describe an airfoil shape with a finite set of variables.

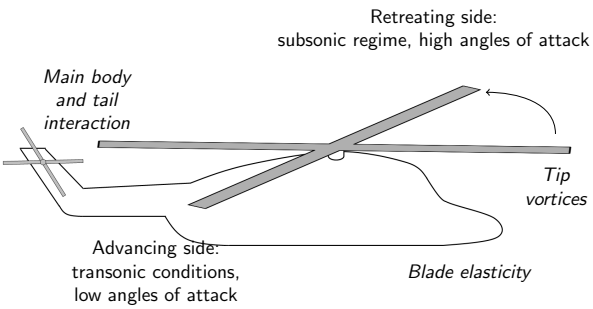


Figure 1: Characteristics of the complex flowfield around a helicopter rotor in forward flight.

The parameterization is well-defined by specifying two functions: a geometry class function C and a shape function S that defines the particular shape of the geometry. In this case, the airfoil shape is decomposed into the camber mean line and the normal thickness distribution. The thickness distribution is taken in perpendicular direction with respect to the camber mean line.

Then, the camber line and thickness distribution are such that the upper surface ζ_u and lower surface ζ_l are obtained applying the thickness perpendicular to the camber line, as follows

$$\begin{aligned} \zeta_u &= \zeta_c + \zeta_t \cos(\epsilon) \\ \zeta_l &= \zeta_c - \zeta_t \cos(\epsilon), \end{aligned} \tag{1}$$

where $\epsilon = \arctan\left(\frac{\mathrm{d}\zeta_c}{\mathrm{d}\chi}\right)$. Please note that also the χ coordinate of the resulting airfoil will be affected by this addition of vectors.

Note that the coordinate of the camber mean line at the trailing edge is set to zero to consider closed trailing edge and zero geometric angle of attack. In addition, the class functions for the camber line C_c and the thickness distribution C_t are defined as follows

$$C_c(\chi) = \chi(1-\chi) \quad C_t(\chi) = \sqrt{\chi}(1-\chi). \tag{2}$$

While the class function for the thickness distribution is the one suggested in Ref. [11] to define a rounded nose distribution close to the leading edge, the class function for the camber presents a linear term to avoid vertical slope of the camber distribution at the leading edge. In addition, the linear term of the class function permits to directly relate the first coefficient of the shape function A_0^c with the slope of the camber line at the leading edge.

The shape function $S(\chi)$ is given by a Bernstein polynomial of order N, whose coefficients represent the design variables $\boldsymbol{x}_s$ of the optimization problem. A convergence study, suggested the employment of 4th order polynomials for each distributions that yields 10 design variables. The order of the polynomial could be chosen based on a convergence analysis of the CST (for further details see Ref. [9]).

In addition to the coefficients describing the shape of the airfoil, the angle of attack can be regarded as a design variable. In fact, it is possible to solve the constrained optimization problem by acting on the airfoil design variables $\boldsymbol{x}_s$ to minimize the drag coefficient and on the angle of attack to track the target lift coefficient inside two nested loops. The underlying idea is an

analogy with the blade collective pitch control: to increase trust it is necessary to act on the collective pitch, which changes the angle of attack of the blade section.

Thus, each airfoil tested in the optimization loop is obtained with a specific set $\boldsymbol{x}_s$, and its performance are evaluated computing the angle of attack α that provides the desired lift coefficient. The nested optimization loops used to implement this procedure are described in detail in Section 2.3. Thus, the resulting set of design variables is $\boldsymbol{x} = \{\boldsymbol{x}_s, \alpha\}$.

2.2 Aerodynamic models

A model describing the aerodynamic load acting on the airfoil is necessary to compute the objective function of the optimization problem. The aerodynamic models used in this optimization problem are different for each side of the blade in order to capture the specific features of the flow in such different operating conditions, while limiting the computational effort.

In the retreating side featuring subsonic condition below the static stall boundary, the XFOIL code [12] is adopted, which is an aerodynamic code with coupled panel and integral boundary layer methods developed for the analysis of subsonic, isolated airfoils [12]. This code is chosen because it provides a fast and sufficiently accurate estimation of the aerodynamic force coefficients for a two-dimensional section in the range of angle of attack considered in this problem. Please note that the highest value of the angle of attack in the range is below the static stall limit for most airfoils.

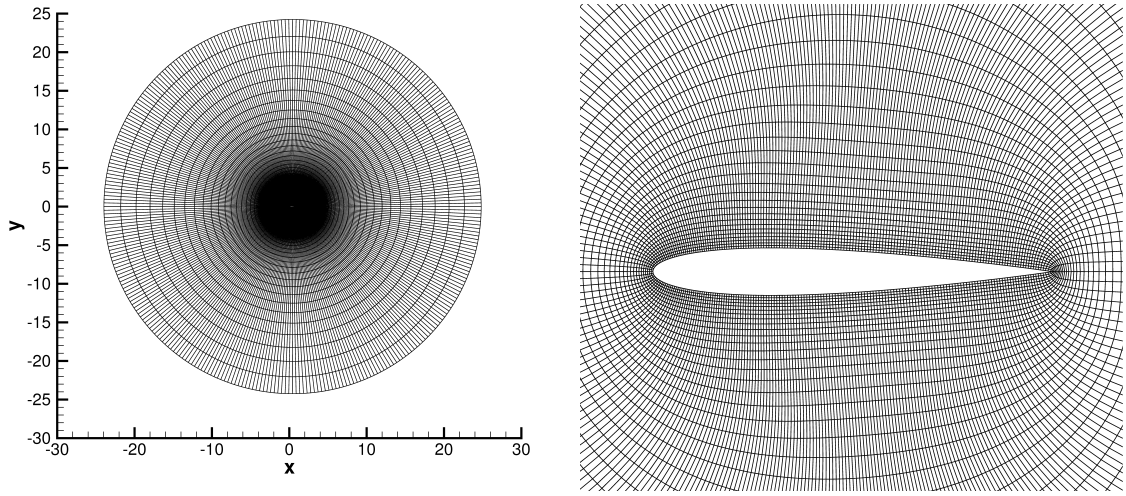
For fully developed transonic conditions, like the one experienced in forward flight, a finite volume-based numerical solver of the Euler equations is used, the Stanford University Unstructured (SU^2) software suite [13, 14]. The SU^2 solver provides several space discretization schemes, among which the second-order Jameson-Schmidt-Turkel scheme is employed. An implicit Euler, local time-stepping is used to converge to the steady-state solution, and the GMRES method in conjunction with the LU SGS preconditioner is used to solve the resulting system [15].

For the computation inside the optimization loop, a two-dimensional coarse structured mesh of 20480 rectangular cells is used (see Fig. 2). The mesh extends ± 25 chords around the unitary-chord airfoil. For each airfoil in the optimization loop, the computation runs for 500 iterations starting from a baseline solution of the NACA0012 airfoil. The mesh is deformed by displacing the airfoil boundary cells according to the new geometry and by translating the displacement to the volume cells with a method based on the finite element method discretization of the linear elasticity equations [13]. The element stiffness is set inversely proportional to the wall distance and 500 smoothing iterations are applied to get a regularly deformed mesh.

2.3 Optimization algorithm

To solve the optimization problem a Non-dominated Sorting Genetic Algorithm [16] is chosen for its ability to explore the design space.

Evolutionary algorithms have been employed to tackle many engineering optimization problems. Applications to aerodynamic optimization problems can be found for instance in Refs. [17–19]. Main advantages of such approaches are the possibility to tackle multi-objective problems without scalarization and to exhaustively explore the design space. On the other hand, they are



(a) O-mesh. (b) Zoom.

Figure 2: Coarse structured grid (20480 cells) for Euler computations inside the optimization loop.

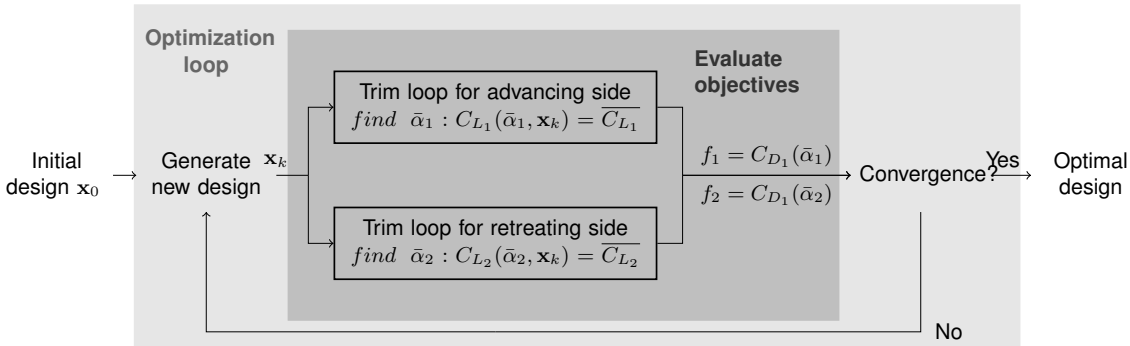


Figure 3: Outer NSGA loop for $\boldsymbol{x}_s$ and inner trim loop to determine $\overline{\alpha}$.

very expensive from a computational point of view because they require many computations of the objective function, even in region of the domain where designs with poorer performance are present. Genetic algorithms uses operators inspired by natural evolution to get to the optimal solution. In such a framework, each design vector is considered as an individual, and the design variables in the vector represent the chromosomes. A group of individuals is called a generation. Each individual is assigned a fitness function, that is a measure of the probability that this individual will be a parent of the individuals of the next generation. The fitness in the optimization algorithms is associated to the objective function: the better the objective function, the higher the fitness value. In the single objective case, the fitness function is a measure of the objective function. In the multi-objective case, the fitness function is assigned based on the position of the individual in the objective space with respect to the current Pareto front. With the fitness value, the selection step is performed to determine the parents of the new generation; mutation and crossover operators are then used to mantain genetic diversity and to obtain different children from a single set of parents. The genetic algorithm performs a loop over the generations. The starting point is an initial generation of individuals that are typically randomly chosen. For each iteration, a new generation of individuals is computed using selection, crossover and mutation operators. Then, the objective functions are evaluated for each individual, and each design vector is assigned a fitness function. From the fitness of these individuals, a new generation is computed and the loop goes on for a sufficient number of generations.

Typical values for the crossover and mutation probabilities $pc = 0.9$, $pm = 0.1$ are chosen and

the sharing parameter is set using a formula based on the number of design variables.

Inside the genetic algorithm loop, an inner loop is nested to deal with the lift coefficient constraint (see Fig. 3). In particular, for each design vector $\boldsymbol{x}_{s,k}$ of the k-th iteration in the genetic loop, a secant method is used to find the angle of attack $\overline{\alpha}$ that guarantees $C_{L,k} = \overline{C_L}$. The i-th iteration of the secant loop reads

$$\alpha_i = \alpha_{l_i} + \frac{\alpha_{r_i} - \alpha_{l_i}}{C_L(\alpha_{b_i}) - C_L(\alpha_{a_i})} \left(\overline{C_L} - C_L(\alpha_{a_i})\right) \tag{3}$$

where α_{l_i} and α_{r_i} are the left and right boundaries of the interval in which the angle of attack is sought and they satisfy: $C_L(\alpha_{r_i}) > \overline{C_L}$ and $C_L(\alpha_{l_i}) < \overline{C_L}$. The left and right value are updated according to the value of the lift coefficient for the current α_i: if $C_L(\alpha_i) > \overline{C_L}$, then $\alpha_{r_{i+1}} = \alpha_i$; if $C_L(\alpha_i) < \overline{C_L}$, $\alpha_{l_{i+1}} = \alpha_i$, and if $C_L(\alpha_i) = \overline{C_L}$ the loop breaks.

The method converges in a few iterations, owing to the quasi-linearity of the lift coefficient for most airfoils in the vicinity of the considered values of the angle of attack. Lack of convergence within few iterations is not critical, because this happens for airfoils with poor values of the lift coefficient that should be discarded anyway. The result of the two nested loops is a set of optimal solutions defined by a design vector $\{\boldsymbol{x}_s, \alpha\}$. It is noted that the inner trim loop targets the equality constraint on the lift coefficient, but the design vector coming out of it may not satisfy the moment coefficient constraint.

3 EFFECTS OF CAMBER AND THICKNESS MORPHING

In this section, the effect of modification of the thickness and the camber of the airfoils is discussed. An airfoil geometry can be characterized by the coordinates of the upper and lower surface. However, an airfoil can also be described by the construction of a thickness shape distributed around a camber line. This description dates back to the 1920s and 1930s, when the NACA 4-digit and 5-digit airfoils were generated by superimposing a simple meanline shape with a thickness distribution that was obtained by fitting a group of popular airfoils of the time.

The camber mean line mainly affects the chordwise load distribution, the angle of zero lift, the pitching-moment coefficient, the slope of the lift curve and the approximate position of the aerodynamic center [20, p.65]. Effects of increasing the camber line are generally a lower zero-lift angle of attack and a higher maximum lift coefficient. The counterpart of this is an increase in the drag coefficient, especially at low angle of attack, and an increase in the pitching moment coefficient. This latter problem, which is crucial in helicopter blade design, led to the almost universal use of symmetric airfoil sections on helicopter developed before the 1960s [21, p.360]. However, with the advent of new camber line distribution, such as the NACA 23012 airfoil which is designed to limit the moment coefficient, cambered airfoils were introduced also in rotorcraft blades. A clear example of the effect of camber on moment coefficient is given by reflexed airfoils, which have reduced camber over the afterward section producing less lift over this region, thereby less nose-down pitching moment. In this case the aft section is actually generating downforce and the moment coefficient at zero lift is positive.

Effects of thickness distribution includes the increase of maximum lift coefficient with increasing thickness, at least for thickness-to-chord ratio in the range used for aeronautical applications (i.e. around 16% [20, p.135] for the NACA 4- and 5-digit airfoils), above which the effect of thickness become detrimental for separation. Thickness distribution has a significant impact

on profile drag and on wave drag. With regard to the latter, earlier drag rise and lift break are experienced for thicker airfoil [20, p.274], although this is in general an effect of both thickness-to-chord ratio and distribution, and may vary from airfoil to airfoil.

3.1 Camber and thickness morphing from an optimal trade-off airfoil

In this section, an ideal morphing strategy is investigated: the assumption is that the morphing mechanism permits to have no geometry constraints to the airfoil shape between the advancing and retreating side, and then each shape is designed by running a shape optimization in one specific operating condition. The interest is to compute an ideal reference gain, with a morphing strategy that could morph the section of a blade from the optimal shape in the advancing side to the optimal shape in the retreating side.

3.1.1 Deterministic optimization

Then, a deterministic optimization of helicopter rotor airfoil in forward flight is performed, which consists typically in the minimization of the drag coefficient C_D ensuring the satisfaction of constraints on the lift C_L and moment C_M coefficients [22–24].

Here, a target lift coefficient $\overline{C_L}$ is set to consider a specific trim condition and the moment coefficient is bounded with a given threshold $\overline{C_M}$ to avoid excessive loading on the blade structure. The objectives of the advancing and retreating side are conflicting with one another. In fact, an optimal shape for the advancing side would be a mildly cambered airfoil to postpone drag rise to higher Mach number values, whereas the adoption of a highly-cambered airfoil in the retreating side could better adapt to higher angle of attack, for instance by means of a nose-droop.

In mathematical terms, two single-objective optimization problems may be formulated as follows

$$\begin{aligned} \text{minimize: } & C_{D_i}(\boldsymbol{x}) \\ \text{subject to: } & C_{L_i}(\boldsymbol{x}) = \overline{C_{L_i}} \quad |C_{M_i}(\boldsymbol{x})| \leq \overline{C_{M_i}} \\ \text{by changing: } & \boldsymbol{x} \end{aligned} \tag{4}$$

where $\boldsymbol{x}$ are the design variables. The subscript i is equal to 1 in the advancing side and 2 in the retreating side. The blade section of the Bo105 rotor [25] at $r = 0.85$ is considered and the operating conditions for advancing and retreating sides are taken from the corresponding "figure-of-eight" in the (M, α)–plane from Ref. [26, p.296]. Both operating Mach number and the corresponding constraints are reported in Tab. 2. The values of the lift constraint $\overline{C_{L_i}}$ are set for each side with the intent of producing a design that improves the performance of the NACA 23012 airfoil, a classical shape for helicopter rotors (the aerodynamic coefficients of the NACA 23012 airfoil in the operating conditions are presented in Tab. 1).

To complete the scenario of deterministic optimization problems, a one-objective optimization with a trade-off performance is performed. The optimization problems in this case is defined as follows

$$\begin{aligned} \text{minimize: } & \frac{1}{2}\sum_{i=1}^{2} w_i C_{D_i}(\boldsymbol{x}) \\ \text{subject to: } & C_{L_i}(\boldsymbol{x}) = \overline{C_{L_i}} \quad |C_{M_i}(\boldsymbol{x})| \leq \overline{C_{M_i}} \;\; i = 1, 2 \\ \text{by changing: } & \boldsymbol{x}, \end{aligned} \tag{5}$$

Case	α[deg]	M[-]	Re [-]	C_L [-]	C_D [counts]	C_M [-]
1.	-1.5	0.75	4.6e6	-0.025	86.97	0.02
2.	12.5	0.28	1.7e6	1.411	191.3	0.03

Table 1: Aerodynamic coefficients of NACA 23012 airfoil.

Case	M[-]	Re [-]	$\overline{C_L}$	$\overline{C_M}$	w_i [1/counts]
1.	0.75	4.6e6	0.025	0.08	1/2
2.	0.28	1.7e6	1.411	0.04	1/160

Table 2: Operating conditions and aerodynamic constraints.

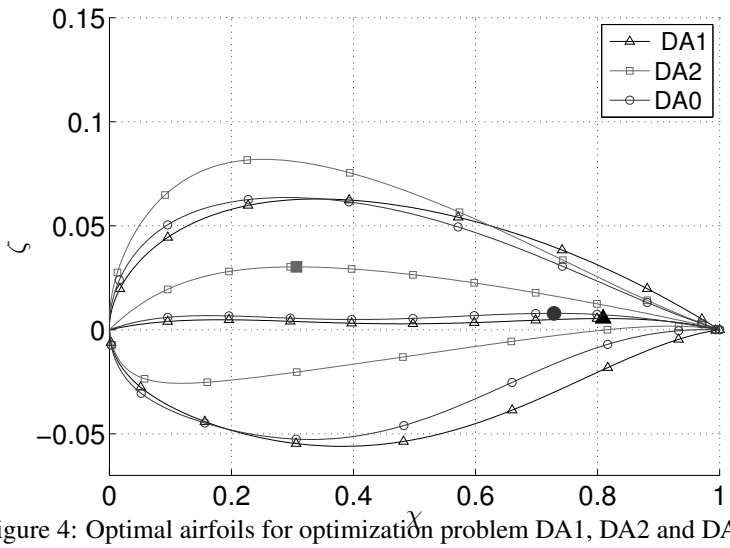


Figure 4: Optimal airfoils for optimization problem DA1, DA2 and DA0.

where w_i are weights chosen from the drag coefficient of the optimal airfoils obtained in the one-point optimization case (see Tab. 2). The result of such an optimization is a compromise solution between advancing and retreating side that is used as a reference of a trade-off optimal solution.

The, three different deterministic optimization problems are solved: (i) airfoil DA1 with optimal drag coefficient in the advancing side, (ii) airfoil DA2 with optimal drag coefficient in the retreating side, and (iii) airfoil DA0 with optimal weighted drag coefficients. These airfoils are presented in Fig. 4 and the characteristics of their geometry is reported in Tab. 3. It is possible to note that the optimal airfoil for the retreating side is highly cambered to cope with the higher angle of attack and it has a lower thickness-to-chord ratio to reduce the drag coefficient. The optimal airfoil of the advancing side generates positive lift owing to a slight camber and greater thickness. The slope of the camber line at the leading edge is controlled by the first coefficient of the parameterization thanks to the modification of the CST discussed in Section 2.1.

The drag coefficients of the optimal airfoils are presented in Tab. 4. The drag coefficient are expressed in drag counts, where one drag count is equal to 0.0001. The gain with respect to the reference NACA 23012 airfoil are presented in Tab. 5. The DA1 airfoil and the DA2 airfoil significantly improve the performance of the advancing and retreating side, respectively.

	DA1	DA2	DA0
Maximum thickness	0.1187	0.1049	0.1161
Position of maximum thickness	0.36	0.22	0.31
Maximum camber	0.0054	0.0303	0.0079
Position of max camber	0.81	0.31	0.73
First design variable (A_0^c)	0.065	0.289	0.107
Camber slope at LE [deg]	3.704	16.098	6.108

Table 3: Characteristics of the optimal airfoils (for unitary chord).

	C_{D_1} [counts]	$\alpha_1\|_{\overline{C_{L_1}}}$ [deg]	C_{M_1} [-]	C_{D_2} [counts]	$\alpha_2\|_{\overline{C_{L_2}}}$ [deg]	C_{M_1} [-]
DA0	0.8947	-1.059	0.0463	177.8	11.1	-0.0108
DA1	0.6927	-1.002	0.0459	–	–	–
DA2	191.99	-3.101	0.1047	126.0	9.27	-0.0393

Table 4: Performance of the optimal airfoils: performance of DA1 in the retreating side is not reported because DA1 airfoil cannot satisfy lift constraint.

	Percentage gain side 1	Percentage gain side 2	Relative gain side 1	Relative gain side 2
DA0	+99%	+7 %	1/97.2	1/1.08
DA1	+99%	–	1/125	–
DA2	–	+34%	–	1/1.52

Table 5: Gain of the optimal airfoils with respect to the NACA 23012 airfoil: gain of DA1 in the retreating side and gain of DA2 in the advancing side are not reported because these airfoils don't satisfy respectively the lift and moment constraint.

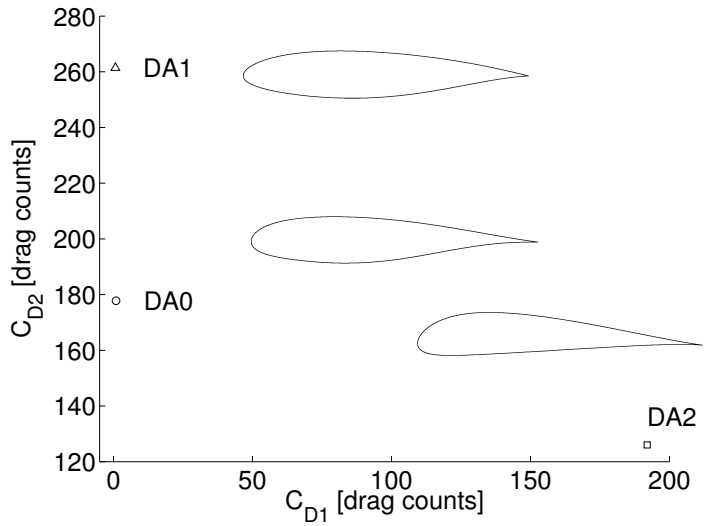


Figure 5: Optimal deterministic shapes and drag coefficient in the advancing and retreating sides

3.1.2 Assessment in terms of morphing strategy

The optimized airfoils are presented in Fig. 5 in the space of the objectives of the deterministic optimization (i.e. the drag coefficient in the advancing and retreating side). Airfoil DA0 repre-

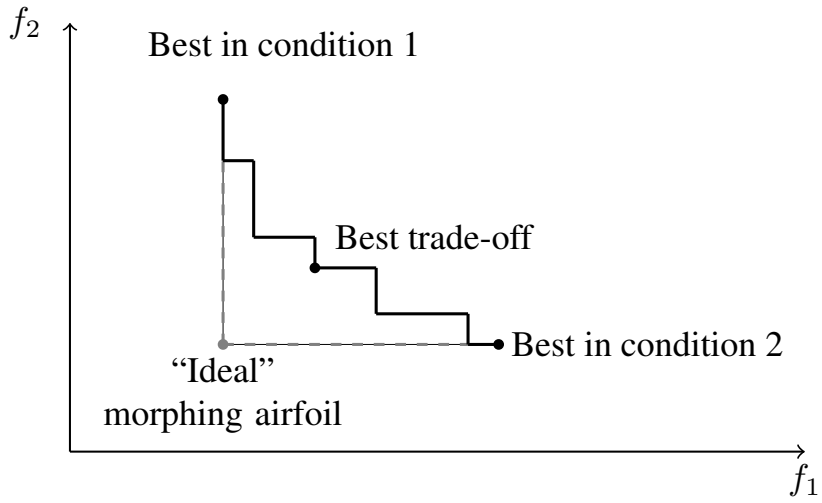


Figure 6: Scheme of two-point optimization and "ideal" morphing airfoil.

	DA1	DA2
Gain w.r.t DA0	22 %	29 %

Table 6: Gain of an "ideal" morphing with respect to the trade-off non-morphing solution DA0.

	Thickness	Camber
Thick DA0 + Camb DA1	DA0	DA1
Thick DA0 + Camb DA2	DA0	DA2
Thick DA1 + Camb DA0	DA1	DA0
Thick DA2 + Camb DA0	DA2	DA0

Table 7: Summary of airfoils obtained by imposition of either the thickness or camber line of DA1 and DA2 from DA0.

sents the reference optimal solution for a non-morphing configuration. Instead, airfoil DA1 and DA2 give an indication of the best possible gain of a morphing strategy. In fact, the maximum, "ideal" gain of morphing airfoils consists of the gain of a morphing strategy that could morph the section of a blade from the shape of airfoil DA1 in the advancing side to the shape of airfoil DA2 in the retreating side. This is depicted in Fig. 6. This solution would have the lowest drag coefficient in the advancing side and at the same time the lowest drag coefficient on the retreating side. The morphing strategy is ideal in this case, because in practice any morphing mechanism would impose geometry constraints to the airfoil shape between the advancing side and retreating side that are not considered in this first instance. Indeed, state-of-the-art technology permits only local deformation of the camber line, as presented in the introduction, although an active field of research on materials is developing ways of increasing the degree of freedom of morphing geometries that could be adopted in future research activities. Nevertheless, the ideal gain is of the greatest significance, because it sets a reference gain. The gain is computed with respect to the optimal non-morphing geometry, which in this case is the trade-off airfoil DA0. As presented in Tab. 6, such a "perfectly" morphing airfoil could gain 22% of the drag coefficient in the advancing side and 29% in the retreating side, while maintaining the trim lift coefficient and moment constraint.

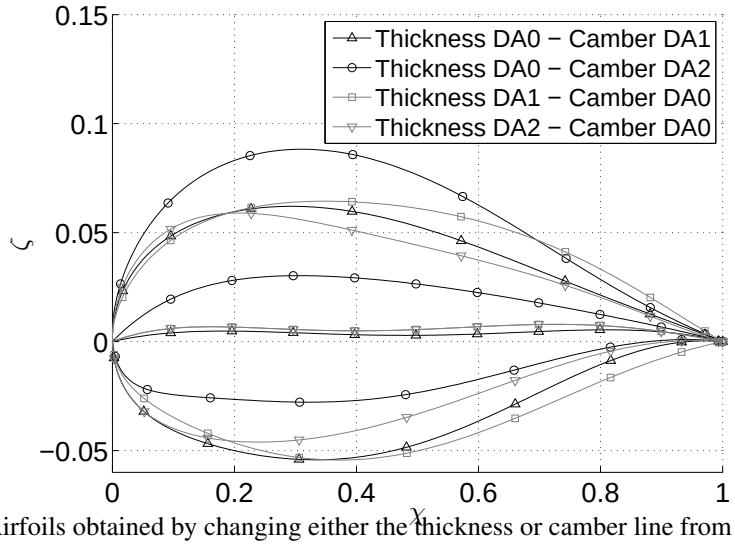


Figure 7: Airfoils obtained by changing either the thickness or camber line from airfoil DA0.

	C_{D_1} [counts]	$\alpha_1\|_{\overline{C_{L_1}}}$ [deg]	C_{D_2} [counts]	$\alpha_2\|_{\overline{C_{L_2}}}$ [deg]	Gain side 1	Gain side 2
DA0	0.8947	-1.059	177.8	11.1		
Thick DA0 + Cam DA1	0.805	-0.945	192.3	11.502	+10%	
Thick DA0 + Cam DA2	151.53	-3.316	141.6	9.55		+20%
Thick DA1 + Cam DA0	0.801	-1.128	–	–	+10.5%	
Thick DA2 + Cam DA0	11.26	-1.028	166.0	10.881		+6%

Table 8: Drag coefficient of airfoils obtained from deterministic optimal airfoils (see Tab. 7).

To further support the choice of camber line morph over thickness distribution morph, a comparison is made among airfoils generated from airfoil DA0 imposing either the thickness distribution of airfoil DA1 and DA2 or the camber line. In other words, four airfoils are constructed by fixing either the thickness or the camber line of airfoil DA0 and imposing the camber line or the thickness of airfoils DA1 and DA2. The airfoils are summarised in Tab. 7 and the resulting shapes are presented in Fig. 7. The first two solutions refers to a camber morphing in which the camber is modified and the thickness is held constant (and equal to the one of the best non-morphing solution). On the other hand, the last two solutions represent a thickness morphing, where the camber is held fixed to the one of airfoil DA0.

For these airfoils the angle of attack for which the lift coefficient satisfies the trim constraint is computed and the corresponding drag coefficients are evaluated. Results are presented in Tab. 8. As explained in Section 2.2, in the advancing side the Euler equations-based CFD model is used and XFOIL in the retreating side (see Sec. 2.2). It is possible to note that the solutions obtained by holding the thickness fixed and changing the camber line (Thick DA0 + Cam DA1 and Thick DA0 + Cam DA2) improves the gain of the reference DA0 airfoil by 10% in the advancing side and 20% in the retreating side. With respect to the ideal gains presented in Tab. 6, this solution restores 50% of the ideal gain in the advancing side and 90% in the retreating side. On the other hand, solutions obtained by modifying the thickness distribution present a similar gain on the advancing side (10.5%), but a smaller gain in the retreating side (6%). These results show that the camber morphing gets closer to the ideal morphing gain and it is therefore more effective. Furthermore, it seems that the thickness and camber line equally contributes to the ideal gain

$\alpha_1\|_{\overline{C_{L_1}}}$ [deg]	$C_{D_1}\|_{\overline{C_{L_1}}}$ [counts]	$C_{M_1}\|_{\overline{C_{L_1}}}$	$\alpha_2\|_{\overline{C_{L_2}}}$ [deg]	$C_{D_2}\|_{\overline{C_{L_2}}}$ [counts]	$C_{M_2}\|_{\overline{C_{L_2}}}$
-3.316	151.53	0.109153	9.55	0.01416	-0.0414

Table 9: Performance of Thick DA0 + Cam DA2 airfoil in both sides.

in the advancing side, while in the retreating side the camber line plays a key role. As a matter of fact, in the advancing side transonic flow is experienced, and the minimization of the drag coefficient corresponds to the minimization of wave drag. This contribution equally depends on camber and thickness distributions over chordline. On the other hand, high angle of attack and subsonic flow are encountered in the retreating side, thus the gain is basically built upon the adaptation of the airfoil to the high angle of attack achieved through camber.

Finally, results for airfoil Thick DA1+Cam DA0 in the retreating side are not presented because this airfoil cannot meet the trim constrained, as it happened with airfoil DA1 (see Tab. 4). In a similar way, airfoil with DA0 thickness and DA2 camber exceeds the moment coefficient in the advancing side as presented in Tab. 9 (similarly to what happened for DA2, please refer to Tab. 4). To some extent, these airfoils obtained by combination of thickness and camber of DA0, DA1 and DA2 inherit some of the characteristics of the original airfoils. However, it is noted that the moment coefficient of the airfoil Thick DA1+Cam DA2 also exceeds the threshold value $\overline{C_{M_2}}$ set in the optimization loop (see Tab. 9).

4 OPTIMAL MORPHING AIRFOILS

The optimization problems tackled in the case of morphing airfoils maintain the same structure of the cases presented in Section 3.1.1. Thus, two optimization problems are performed, one for the advancing side and one for the retreating side, which seeks the design with minimal drag coefficient with an equality constraint on the lift coefficient and an inequality constraint on the moment coefficient. The same values of target lift coefficients, bounding moment coefficients and operating conditions are used (please refer to Tab. 2).

The difference in this case is that the optimization is performed by acting on the design variables describing the morphing geometry. The morphing design variables are described in the following section.

4.1 Morphing strategy and parameterization

The morphing strategy considered in this work is a conformable camber airfoil, which changes over the period of rotation of the blade. Several technological solutions can be employed to achieve this goal: for instance, the FishBone Active Camber [4], the controllable camber presented in Ref. [2] and chiral structures as the one developed in Ref. [6]. The basic idea in the definition of a camber morphing airfoil is that the thickness distribution is held fixed, while the camber is allowed to change its shape during flight.

The type of morphing introduced to some extent in the previous section permits modification of the camber over the entire chordwise coordinate. However, in general the camber line cannot be modified at any chordwise coordinate, but it is held fixed in particular areas to maintain the internal structure of the blade and to accommodate morphing mechanisms. In general, the region

close to the leading edge is fixed, where the D-shaped spar used in the helicopter blade structure is found. Thus, two types of camber morphing are considered. In the first case (Fig. 8a), complete freedom is given to the camber line. In the second type (Fig. 8b) the first part of the airfoil is fixed, while for the remaining part of the blade section, the camber can morph and work as a trailing-edge flap with a larger extension and continuous shape modification. In this way, an increasing level of constraints to the morphing strategy are given: (i) ideal morphing, with constraints neither on the thickness nor on the camber line, (ii) camber morphing, with a fixed thickness distribution, and (iii) trailing edge morphing, with a fixed thickness distribution and a partial constraint on the camber line.

A suited parameterization is required to describe and optimize a morphing camber airfoil. The parameterization introduced in Sec. 2.1 treats separately the camber line and the thickness distribution perpendicular to the local camber line. For the entire camber morphing, the design variables are the coefficients A_i^c of the shape function $S_c(\chi)$ used to describe the camber distribution as $\zeta_c(\chi) = C_c(\chi)S_c(\chi)$. The class function is the same function presented in Eq. (2). The order of the Bernstein polynomials is retained from the optimization in forward flight condition, but owing to the fact that the thickness distribution is now fixed, the number of design variables is reduced by half. It follows that the design variables for the case of camber morphing are $\boldsymbol{x}_{m,c} = \{A_i^c\},\ i = 0, ..., 4$.

For the trailing edge morphing modification, a piecewise cubic function is used. The cubic functions are defined to ensure continuity up to the second order. Following an approach similar to the one presented in Ref. [10], the morphing camber line is described as follows

$$\begin{aligned} \zeta_c(\chi) &= C_c(\chi)\ S_c(\chi) + H(\chi) \\ H(\chi) &= \sum_{i=1}^{n_c} h_i(\chi) \end{aligned} \tag{6}$$

and each cubic function $h_i(\chi)$ reads

$$h_i(\chi) = \begin{cases} 0, & \text{if } \chi \leq \psi_{0,i} \\ A_i^m \frac{(x-x_{0,i})^3}{(1-x_{0,i})^3}, & \text{if } \chi > \chi_{0,i}, \end{cases}$$

where a_i are the coefficients of the cubic functions and $x_{0,i}$ are the locations of the start of the morph. The number of cubic functions n_c is equal to two in order to represent also reflex cambered airfoil and the locations of the morphs are $x_{0,1} = 0.4$ and $x_{0,2} = 0.7$. The D-spar length $x_{0,1}$ is set to 40% of the chord, following the FishBAC configuration [10]. The coefficients of the cubic functions are the design variables of the morphing airfoil $\boldsymbol{x}_{m,te} = \{A_1^m, A_2^m\}$.

(a) Camber line morphing. (b) Trailing edge morphing.

Figure 8: Types of morphing strategy: baseline geometry (black), camber morph (green) or trailing edge morph (blue) and resulting morphed geometry (red) .

4.2 Deterministic results

With two different operating conditions and two different morphing strategies, four optimization problems are performed, which are summarised in Tab. 10. Each optimization has been performed with an initial generation of 100 individuals and 10 generations of 55 individuals to widely explore the solution space. The results of these optimization are reported in Tab. 11, where the first and second columns indicate the morphing strategy and the objective, respectively. The last two columns present the gain with respect to the optimal non-morphing airfoil, i.e. airfoil DA0.

The camber morphing involving the entire camber line permits to gain 23% with respect to DA0 in the retreating side and 10% in the advancing side. In the latter condition, the optimal solution obtained by the optimization is indeed the airfoil obtained by summing the thickness distribution of DA0 to the camber line of airfoil DA1, i.e. Thick DA0+Cam DA1. In the retreating side, the optimization is capable of improving the drag coefficient of airfoil Thick DA0+Cam DA2 and at the same time overcoming the problem associated with the moment coefficient constraint.

On the other hand, the trailing edge morphing improves the starting airfoil of 5% in the retreating side, but it does not find better solutions than airfoil DA0 in the advancing side. A first explanation of this can be found in the fact that airfoil DA0 has a shape more akin to DA1 than DA2 (see Fig. 5) and it is an airfoil that already greatly reduces the wave drag coefficient of the reference airfoil (i.e. the NACA 23012 airfoil shown in Tab. 1). However, the modification that the camber needs to undergo to improve the performance in the advancing side (for example, a modification into the camber line of airfoil DA1) is not compatible with the trailing edge modification shown in Fig. 8b. This problem could be overcome only by changing the thickness distribution together with the trailing edge modification. The best gain of such a strategy may be defined only by an optimization problem combining the coefficient of the thickness distribution A_i^t, the coefficient of the mean line A_i^c and the coefficient of the local trailing edge morph A_i^m.

To appreciate the difference in the morphing strategies, the optimal airfoils for the retreating side are presented in Fig. 9. It is worth noting how the camber morphing lifts up the camber line while maintaining the trailing edge location, and the trailing edge morph deflects the trailing edge to achieve an increase in lift coefficient.

4.3 Robust results

From the deterministic optimization, valuable information has been obtained. In fact, the camber morphing could potentially bridge the gap between a non-morphing airfoil and an "ideal" morphing airfoil and the morphing over the entire chordwise distribution of the camber line is

Case	Objective	Equality constraints	Inequality constraints	Design variables
DMO1	$\min C_{D,1}$	$C_{L_1} = \overline{C_{L_1}}$	$\lvert C_{M_1}\rvert \leq \overline{C_{M_2}}$	$\boldsymbol{x}_{m,c}, \alpha$
DMO2	$\min C_{D,2}$	$C_{L_2} = \overline{C_{L_2}}$	$\lvert C_{M_2}\rvert \leq \overline{C_{M_2}}$	$\boldsymbol{x}_{m,c}, \alpha$
DTO1	$\min C_{D,1}$	$C_{L_1} = \overline{C_{L_1}}$	$\lvert C_{M_1}\rvert \leq \overline{C_{M_2}}$	$\boldsymbol{x}_{m,te}, \alpha$
DTO2	$\min C_{D,2}$	$C_{L_2} = \overline{C_{L_2}}$	$\lvert C_{M_2}\rvert \leq \overline{C_{M_2}}$	$\boldsymbol{x}_{m,te}, \alpha$

Table 10: Summary of optimization problems.

Morphing	Objective	C_{D_1} [counts]	$\alpha_1\|_{\overline{C_{L_1}}}$ [deg]	C_{D_2} [counts]	$\alpha_2\|_{\overline{C_{L_2}}}$ [deg]	Gain side 1	Gain side 2
Camber	$\min f_1$	0.805	-0.945	192.3	11.502	+10%	
Camber	$\min f_2$	102.82	-2.5	135.7	9.62		+23%
TE morph	$\min f_1$	0.8947	-1.059	177.8	11.1	0%	
TE morph	$\min f_2$	100.50	-1.93	169.2	10.045		+5%

Table 11: Performance of the deterministic airfoils obtained with two strategies for camber morphing

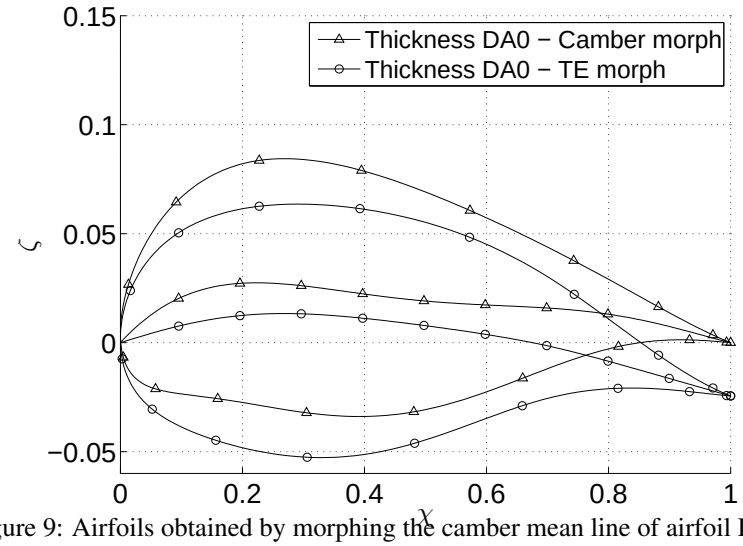


Figure 9: Airfoils obtained by morphing the camber mean line of airfoil DA0.

more effective to this purpose. Starting from these results, the robust optimization problems are tackled in this section. Note that the design variables used in this case are the design variable of the camber morphing $\boldsymbol{x}_{m,c}$, as this technique is more effective.

The objective of robust optimization is to design an airfoil that is minimally sensitive to the variation of the operating conditions. In this case, it is considered that the operating conditions at which the blade section will operate are affected by the uncertainty arising due to modelling assumptions of physical parameters necessary at the design stage.

4.3.1 Robust optimization formulation

Due to the uncertainty, the objective function f is no longer only a function of the design variables $\boldsymbol{x}_{m,c}$, but it also depends on the uncertain variables $\boldsymbol{\xi}$. Within this uncertain framework a minimization with a target lift coefficient in a specific design conditions loses its meaning. However, a robustly optimal airfoil with a satisfactory lift-to-drag ratio in a range of conditions could be used to trim the aircraft in a specific condition keeping always a low drag coefficient. In addition, if the robustness of the lift-to-drag ratio will translate into the drag coefficient at a specific target lift coefficient, a robust shape would ensure less variability of the required power due to aerodynamic drag.

In particular, for the problem under analysis, the following objectives are defined for each side of the blade.

1. On the advancing side, rotor blade encounters transonic flow, so airfoils should typically be designed to delay drag divergence to higher Mach numbers [21, 27]: however, since

the airfoil should feature also a certain amount of lift, the lift-to-drag ratio is maximized for the first condition corresponding to the advancing blade at 90 deg, with the constraint of providing a lift coefficient equal to or greater than the value of the reference NACA 23012 airfoil. Thus, the lift-to-drag ratio C_L/C_D is the objective f_1 of the optimization: $f_1 = C_L/C_D$.

2. On the retreating side, the increase of the lift coefficient is typically sought [27]: for the blade at 180 deg the ratio $C_L^{3/2}/C_D$ is maximized, which is a measure related to the rotor figure of merit [21] and privileges the lift coefficient over the drag coefficient, that is $f_2 = C_L^{3/2}/C_D$.

The nominal operating conditions of the aerodynamic simulation are set using data from Ref. [26]. Finally, the constraints on the moment coefficient used in the deterministic case (and presented in Tab. 2) are still considered.

In mathematical terms, the resulting optimization problem can be stated as:

$$\begin{aligned} \text{maximize :} \quad & \mu\left(f_i\left(\boldsymbol{x}_s, \boldsymbol{\xi}\right)\right) \\ \text{and minimize :} \quad & \sigma^2\left(f_i\left(\boldsymbol{x}_s, \boldsymbol{\xi}\right)\right) \\ \text{subject to :} \quad & \mathbf{g}\left(\boldsymbol{x}_s, \boldsymbol{\xi}\right) \leq \mathbf{0} \\ \text{by changing :} \quad & \boldsymbol{x}_s, \end{aligned} \tag{7}$$

where the moment constraints are collected in vector $\mathbf{g}$. Similarly to the deterministic case, function f_i is equal to f_1 for the advancing side and equal to f_2 in the retreating case. The design variables are the variables defining the shape of the airfoil, while the angle of attack is considered as one of the uncertain variables in the robust design problem.

Indeed, the angle of attack α and the Mach number M encountered by the two-dimensional section of the blade are considered as uncertain, because they are affected by uncertainty on the modeling of the physical parameters considered in the design stage. For instance, both aerodynamic and structural uncertainties, such as blade chord, air density, and rotor angular velocity, may affect the value of the angle of attack and Mach number. Following a probabilistic framework, the uncertain variables are modeled as uniformly distributed random variables around a nominal value. The nominal conditions are reported in Tab. 1 and the uncertainty band is set to 5% for the Mach number and for the angle of attack.

The objective functions of Eq. (7) are the statistics of the performance f. These are computed by means of an uncertainty propagation technique, that is a method to propagate the uncertainty affecting the operating conditions into the quantity of interest f.

4.3.2 Uncertainty quantification technique

A probabilistic framework is employed to deal with the uncertain input data. In this context the stochastic input quantities $\boldsymbol{\xi}$ are treated as independent continous random variables. The random vector $\boldsymbol{\xi}$, whose dimension is equal to the number of uncertain variables n_ξ, belongs to the probability space $(\Omega, \mathcal{F}, P)$, composed by the sample space Ω, the σ-algebra $\mathcal{F}$ of the subsets of the events and a probability measure P. It essentially maps the samples in $\Omega = [0,1]^{n_\xi}$ into the random outcomes $\boldsymbol{\xi} \in \Xi$ and it is characterized by the probability density function $p_\xi(\boldsymbol{\xi})$. The output of the system is then a stochastic variable, and therefore the performance $E(\mathbf{y}, \boldsymbol{\xi})$, which is a function of the solution, is a stochastic variable as well. In the robust

optimization procedure the interest is in reconstructing the mean value and the variance of the quantity of interest; in the following, the stochastic output variable is referred to as $u(\mathbf{y}, \boldsymbol{\xi})$, while its deterministic realization is $\tilde{u}_i(\mathbf{y}, \boldsymbol{\xi}_i)$. Let us define the expected value of the stochastic variable u as follows

$$\langle u \rangle = \int_{\Xi} u(\mathbf{y}, \boldsymbol{\xi})\, p_{\xi}(\boldsymbol{\xi})\, \mathrm{d}\xi,$$

with $p_{\xi}(\boldsymbol{\xi}) = \prod_i^{n_\xi} p_{\xi_i}(\xi_i)$ the joint probability of the independent input variables, and the inner product operator of two stochastic variable u and v with respect to the joint probability (i.e. the covariance for independent u and v)

$$\langle u, v \rangle = \int_{\Xi} u(\mathbf{y}, \boldsymbol{\xi}) v(\mathbf{y}, \boldsymbol{\xi})\, p_{\xi}(\boldsymbol{\xi})\, \mathrm{d}\xi.$$

The mean and variance then read

$$\begin{aligned} \mu(u) &= \langle u \rangle \\ \sigma^2(u) &= \langle u, u \rangle - \langle u \rangle^2. \end{aligned} \tag{8}$$

To reconstruct the quantities in Eq. (8), a Polynomial Chaos (PC) expansion method [28] is used. In fact, under specific conditions, a stochastic process can be expressed as a spectral expansion based on suitable orthogonal polynomials, with weights associated to a particular probability density function. The first study in this field is the Wiener process that was later developed in Ref. [28]. The basic idea is to project the variables of the problem onto a stochastic space spanned by a complete set of orthogonal polynomials Ψ that are functions of the random variables $\boldsymbol{\xi}$. For example, variable u has the following spectral representation:

$$u\left(\mathbf{y}, \boldsymbol{\xi}\right) = \sum_{k=0}^{\infty} \alpha_k\left(\mathbf{y}\right) \Psi_k\left(\boldsymbol{\xi}\right). \tag{9}$$

where Ψ_k are the PC orthogonal polynomials and α_k the coefficients of the expansion. In practice, the series in Eq. (9) has to be truncated to a finite number of terms N, which is determined by

$$N + 1 = \frac{(n_\xi + n_o)!}{n_\xi!\; n_o!},$$

where n_ξ is the dimension of the uncertainty vector $\boldsymbol{\xi}$ and n_o is the order of the univariate polynomial expansion $\phi_i(\xi_i)$ from which the multivariate polynomials $\Psi_k(\boldsymbol{\xi})$ are obtained via tensorization, i.e.

$$\Psi_k\left(\boldsymbol{\xi}\right) = \prod_i^{n_\xi} \phi_i(\xi_i).$$

The polynomial basis $\phi_i(\xi_i)$ is chosen according to the Wiener-Askey scheme [28] in order to select orthogonal polynomials with respect to the probability density function $p_\xi(\boldsymbol{\xi})$ of the input. In this work, because a uniform distribution is considered, Legendre polynomials are employed. The orthogonality property can be advantageously used to compute the PC coefficients of the expansion α_k in a non-intrusive PC framework; this procedure is called Non-Intrusive Spectral Projection (NISP) [17]. In fact, from the orthogonality property it directly follows that

$$\alpha_k = \frac{\langle u(\mathbf{y}, \boldsymbol{\xi}), \Psi_k(\boldsymbol{\xi}) \rangle}{\langle \Psi_k(\boldsymbol{\xi}), \Psi_k(\boldsymbol{\xi}) \rangle} \quad \forall k. \tag{10}$$

Case	Objective	Inequality constraints	Design variables
RMO1	$\max(\mu(f_1))$ and $\min(\sigma^2(f_1))$	$\lvert C_M \rvert \leq \overline{C_M}$	$\boldsymbol{x}_{m,c}$
RMO2	$\max(\mu(f_2))$ and $\min(\sigma^2(f_2))$	$\lvert C_M \rvert \leq \overline{C_M}$	$\boldsymbol{x}_{m,c}$

Table 12: Summary of robust optimization problems.

The computation of the PC coefficients requires an integration of the polynomials that can be estimated with several approaches, among which quadrature formula are chosen in this study. As a result, the solution of the deterministic problem is required for each quadrature point. Once the polynomial chaos and the associated α_k coefficients are computed, the mean value and the variance of the stochastic solution $u(\mathbf{x}, \boldsymbol{\xi})$ are obtained leveraging the orthogonality of the polynomials Ψ_k with respect to the probability function p_ξ:

$$\begin{aligned}
\mu(u)|_{PC} &= \left\langle \sum_{k=0}^{N} \alpha_k(\mathbf{y})\, \Psi_k(\boldsymbol{\xi}) \right\rangle = \alpha_0(\mathbf{y}) \\
\sigma^2(u)|_{PC} &= \left\langle \left(\sum_{k=0}^{N} \alpha_k(\mathbf{y})\, \Psi_k(\boldsymbol{\xi}) \right)^2 \right\rangle - \alpha_0^2(\mathbf{y}) \\
&= \sum_{k=1}^{N} \alpha_k^2(\mathbf{y}) \left\langle \Psi_k^2 \right\rangle .
\end{aligned}$$

Because a uniform distribution is considered also in the forward flight case, Legendre polynomials are employed in the PC expansion. The order of the expansion is retained from a convergence analysis which has been performed and proved that a fourth-order polynomial is sufficient to accurately capture the statistics. The reconstruction of the statistics is based on the evaluation of the exact function for a set of samples $\boldsymbol{\xi}_k$ in the uncertain variables domain for each design vector $\overline{\boldsymbol{x}_s}$.

4.3.3 Results

Two optimization problems are then performed, which are summarised in Tab. 12.

Given the analysis of the previous sections, the nominal operating conditions are set from the deterministic results. In other words, the nominal operating condition of the robust optimization in the advancing side is defined by the angle of attack $\overline{\alpha_1}$ of airfoil DMA1, while for the retreating side the nominal angle of attack is the angle $\overline{\alpha_2}$ of airfoil DMA2.

In the robust optimization problem (RMO2) the Pareto front shown in Fig. 10a is obtained. To select an interesting airfoil from the Pareto set, the lift coefficient obtained in the samples over the stochastic domain are plotted against the target lift coefficient and an airfoil is selected from the front. The selected airfoil is plotted in Fig. 10b together with the deterministic airfoil DMA2. This airfoil is referred to as RMA2 and its performance in the uncertainty range around the operating condition satisfying the lift constraint are reported in Tab. 13. It is possible to note that airfoils DMA2 and RMA2 are very similar and therefore they present similar performance. In this case, the result from the deterministic optimization is a robust airfoil in itself. However, it is important to note that this in general may not be the case and that robust optimization

Airfoil	$C_{D,2}\|_{\overline{C_{L_2}}}$ [counts]	$\alpha_1\|_{\overline{C_{L_2}}}$ [deg]	$\mu_{C_{D,2}}$ [counts]	$\sigma^2_{C_{D,2}}$ [counts^2]	$(\sigma/\mu)_{C_{D,2}}$ [-]
DMA2	135.6	9.62	135.9	39.99	0.047
RMA2	136.6	9.62	136.7	40.06	0.046

Table 13: Drag coefficient of the deterministic airfoil minimizing C_{D_2} (DMA2) and the airfoil selected from the robust front in the retreating side (RMA2).

Airfoil	$C_{D,1}\|_{\overline{C_{L_1}}}$ [counts]	$\alpha_1\|_{\overline{C_{L_1}}}$ [deg]	$\mu_{C_{D,1}}$ [counts]	$\sigma^2_{C_{D,1}}$ [counts^2]	$(\sigma/\mu)_{C_{D,1}}$ [-]
DMA1	0.805	-0.945	3.99	31.09	1.4
RMA1	0.827	-0.894	5.07	43.57	1.3

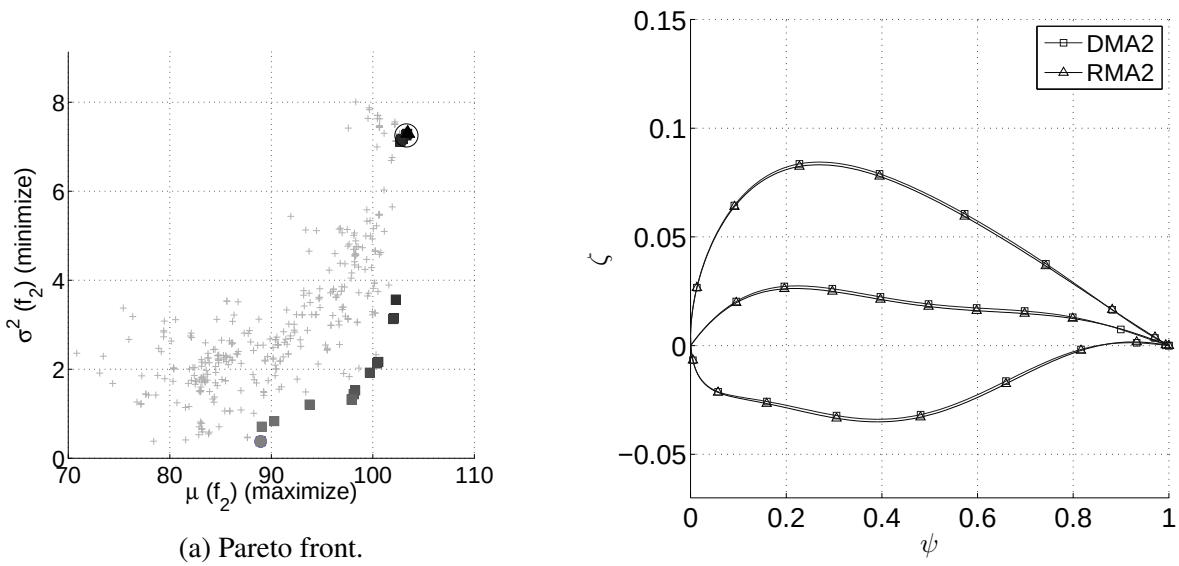
Table 14: Drag coefficient of the deterministic airfoil minimizing C_{D_1} (DMA1) and the airfoil selected from the robust front in the advancing side (RMA1) with camber morphing.

formulation represents a procedure to guarantee that the optimal airfoils would not incur in lift loss, drag increase or moment penalties throughout the uncertainty range.

The results of the robust optimization problem in the advancing side (RMO1) are shown in Fig. 11a. In the Pareto front, airfoil RMA1 is hihglighted which is chosen for its significantly low variance and average mean performance. The shape of airfoil RMA1 is presented in Fig. 11b along with the deterministic optimal airfoil. It is possible to note that even in this case the deterministic solution is very similar to a non-dominated robust airfoil. The deterministic airfoil is also very robust, as demonstrated by the values of the mean and variance presented in Tab. 14.

5 VALIDATION WITH AZIMUTH

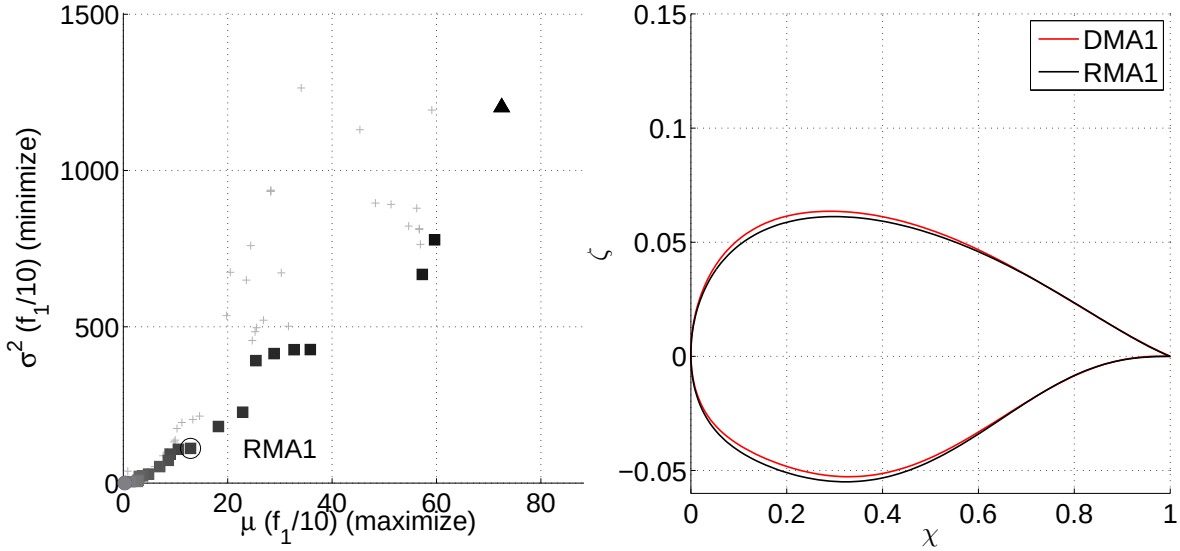
After evaluating the performance in the advancing and retreating side, the drag coefficient of the optimal airfoils over the azimuth angle of the blade rotation is computed. In the case of a



(a) Pareto front.

(b) Optimal deterministic and robust airfoils.

Figure 10: Results of the robust optimization for the retreating side (RMO2) .



(a) Pareto front. (b) Optimal deterministic and robust airfoils.

Figure 11: Results of the robust optimization for the advancing side (RMO1) .

morphing airfoil, a smooth transition from the optimal shape in the advancing side to the optimal shape in the retreating side is required. The transition over the azimuth angle ψ is obtained as follows

$$\zeta = \frac{\zeta_1}{2}(1 + \sin(\psi)) + \frac{\zeta_2}{2}(1 - \sin(\psi)) \tag{11}$$

where ζ_1 is the shape in the advancing side ($\zeta(90°) = \zeta_1$) and ζ_2 is the shape in the retreating side ($\zeta(180°) = \zeta_2$). The transition in the case of the "ideal" morphing DA1-DA2 is presented in Fig. 12, while the morphing shapes with geometrical constraints are shown in Fig. 13. The drag coefficient is then evaluated as follows:

- for the advancing side, an Euler equations simulation with SU^2 is performed, with the set-up used in the optimization loop (Sec. 2.2). Because the Euler equations are used, a correction associated with viscous effects is added to the inviscid SU^2 estimate combining the van Driest II method and a form-factor correction as presented in Ref. [29];
- for the retreating side, the XFOIL solver is adopted
- for the remaining intermediate points, the MSES code is used

The values of the mean angle of attack and the first harmonic amplitude are computed and

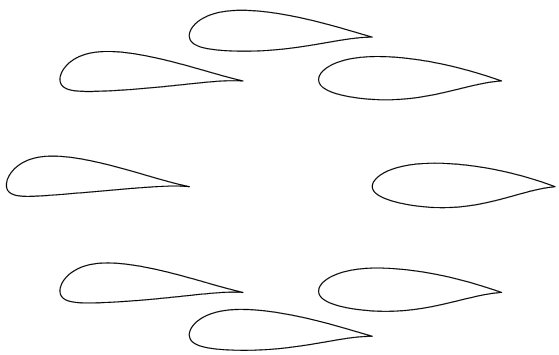
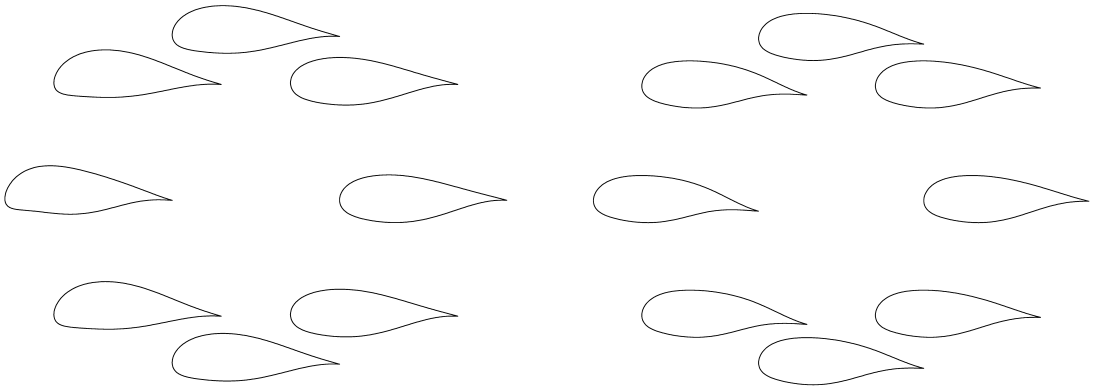


Figure 12: Shapes over the azimuth obtained by morphing DA1 into DA2.



(a) Morphing DMA1 into DMA2. (b) Morphing DTA1 into DTA2.

Figure 13: Shapes over the azimuth.

reported in Tab. 15 for the following combinations: (i) DA1-DA2, i.e. airfoil DA1 morphing into DA2, (ii) DMA1-DMA2, i.e. airfoil DMA1 morphing into DMA2, and (iii) DTA1-DTA2, i.e. airfoil DTA1 morphing into DTA2. The lift and drag coefficients for these airfoils are presented in Fig. 14. It is possible to note that the greatest difference among these airfoils is the performance in the retreting side, where the lowest drag coefficient is obtained for the "ideal" morphing and the drag coefficient increases with increasing geometrical constraints. The performance in the advancing side is comparable among the different morphing strategies, because the viscous correction [29] is dominant with respect to the inviscid drag coefficient.

The average drag coefficient obtained with the "ideal" morphing and the camber morphing are presented in Tab. 16. The table also includes the gain with respect to the non-morphing optimal airfoil DA0 which is higher for the ideal case (22.6 %) and reduces to 17.1% and 3 % in the case of the camber morphing and trailing edge modification, respectively. It is clear that to obtain the maximum gain of the morphing strategies, an optimization which can model the thickness distribution as well is necessary.

6 CONCLUSIONS

A methodology for robust optimization has been developed to tackle the problem of designing morphing airfoils of helicopter rotor blades. Robust optimization requires the coupling of the uncertainty quantification method and the optimization algorithm. This coupling increases the computational demand, especially in aerodynamic applications, where a single function evaluation may be very expensive.

Morphing airfoils have been considered as a possible solution to cope with the variable flow conditions encountered by blade element in forward flight. Deterministic estimates prove that an "ideal" morphing can increase the performance of an optimal non-morphing airfoils by 22.6 % on the average drag coefficient, and by 17 % when thickness constraint is considered.

	DA1-DA2	DMA1-DMA2	DTA1-DTA2
α_0 [deg]	4.13	4.34	4.49
Δ_1 [deg]	4.95	5.10	5.36

Table 15: Variables of time-varying angle of attack for optimal shapes.

	DA1-DA2	DMA1-DMA2	DTA1-DTA2
$avg(C_D)$ [counts]	76.44	81.84	95.14
Gain wrt DA0	22.6 %	17.1 %	3.6 %

Table 16: Average drag coefficient over the blade rotation period for morphing airfoils and associated gain.

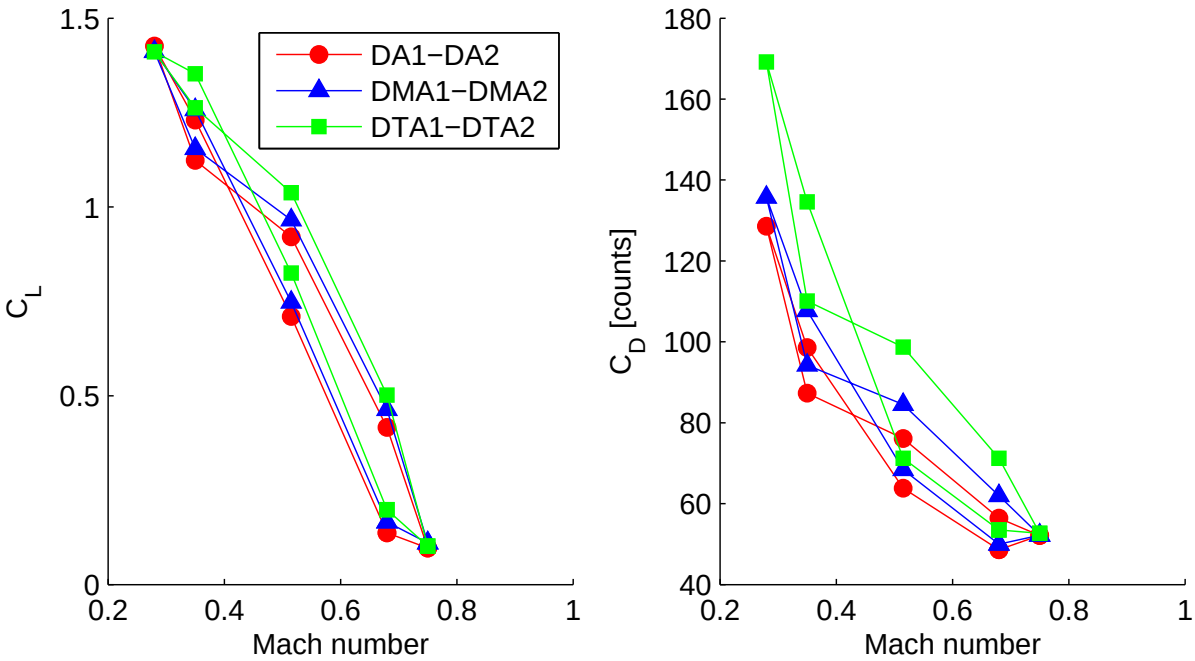


Figure 14: Lift and drag coefficient loop of DA1-DA2, DMA1-DMA2 and DTA1-DTA2.

The employment of robust optimization is thus expected to be of great interest in the application of helicopter rotor airfoils, owing to the fact that even the airfoils with higher mean values achieve lower values of the variance of the aerodynamic efficiency with respect to variations of the angle of attack. The reduction of the variance could lead for instance to a reduction of the required rotor shaft torque in variable operating conditions. Furthermore, in the different applications presented in this paper, results have demonstrated how the robust optimization formulation represents a formal procedure to guarantee that the optimal airfoils would not incur in lift loss, drag increase or moment penalties throughout the uncertainty range.

Future works will be devoted to the unsteady assessment of the morphing geometries to ultimately verify the result of the steady optimization. In addition, an optimization including the thickness distribution as a design variable will be considered for the morphing cases. With regard to the robust technique developed in the thesis, the joint employment of the multi-fidelity method and coupled adaptive approach will be explored to further improve numerical efficiency.

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